

## Appendix II to Lecture 2.2

# Solving and Applying Kepler's Equation

# Numerical Solution of Kepler's Equation

- OK, So How we Extract Numerical Solutions of

## Kepler's Revenge!



Kepler

$$M_{t-0} = \{ E_t - e \sin (E_t) \}$$

## Anatomy of the Solver Algorithm

$$E^{(j+1)} = E^{(j)} + \frac{M - [E^{(j)} - e \sin(E^{(j)})]}{1 - e \cos(E^{(j)})}$$

*Iteration index* (points to  $E^{(j+1)}$ )

*True Mean Anomaly* (points to  $M$ )

*Current estimate of mean anomaly* (points to  $E^{(j)}$ )

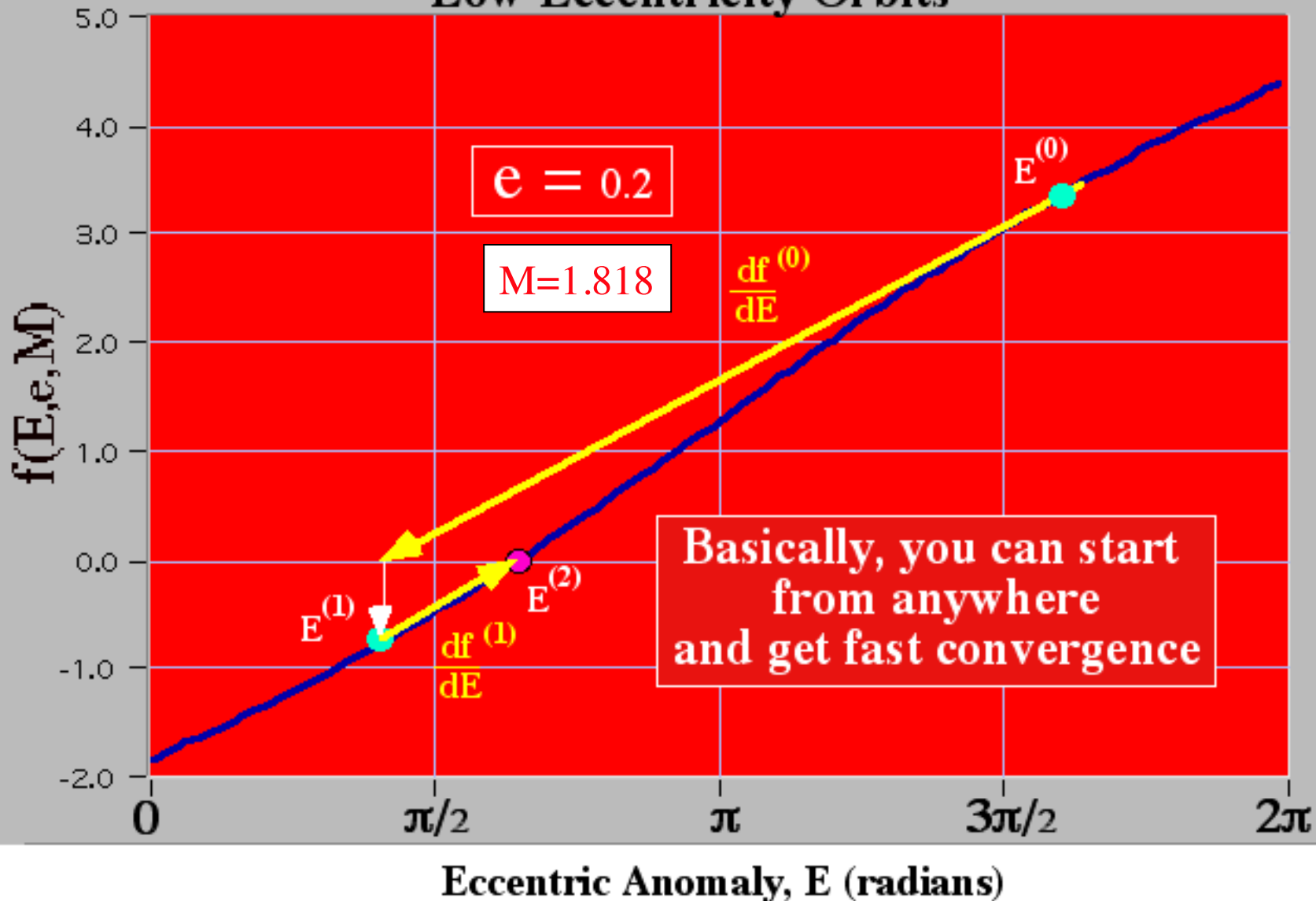
*Current estimate* (points to  $E^{(j)}$ )

*Refined estimate* (points to  $E^{(j+1)}$ )

*Steepest descent* (points to the denominator  $1 - e \cos(E^{(j)})$ )

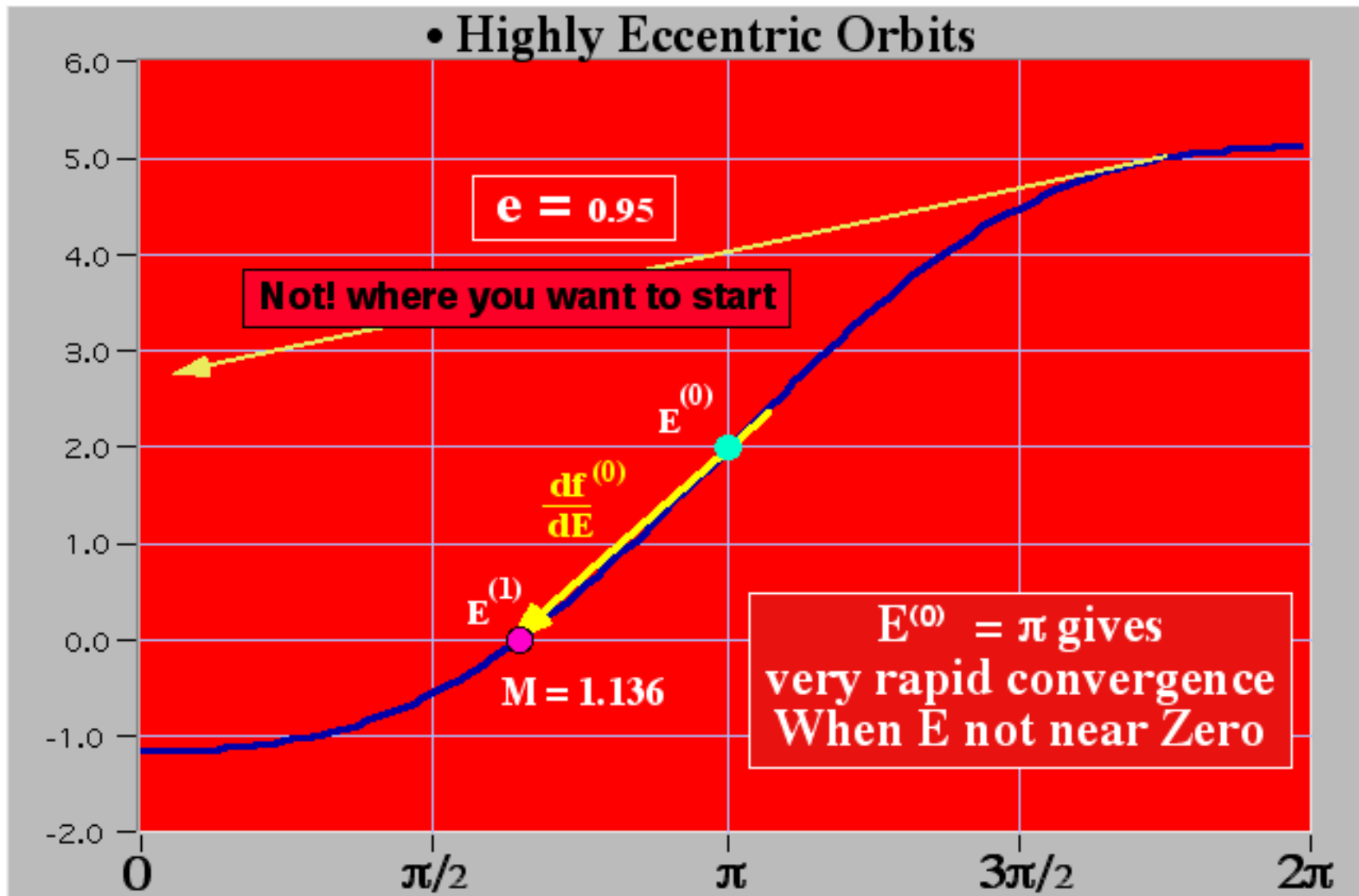
# Starting Value

• Low Eccentricity Orbits



# Starting Value (cont'd)

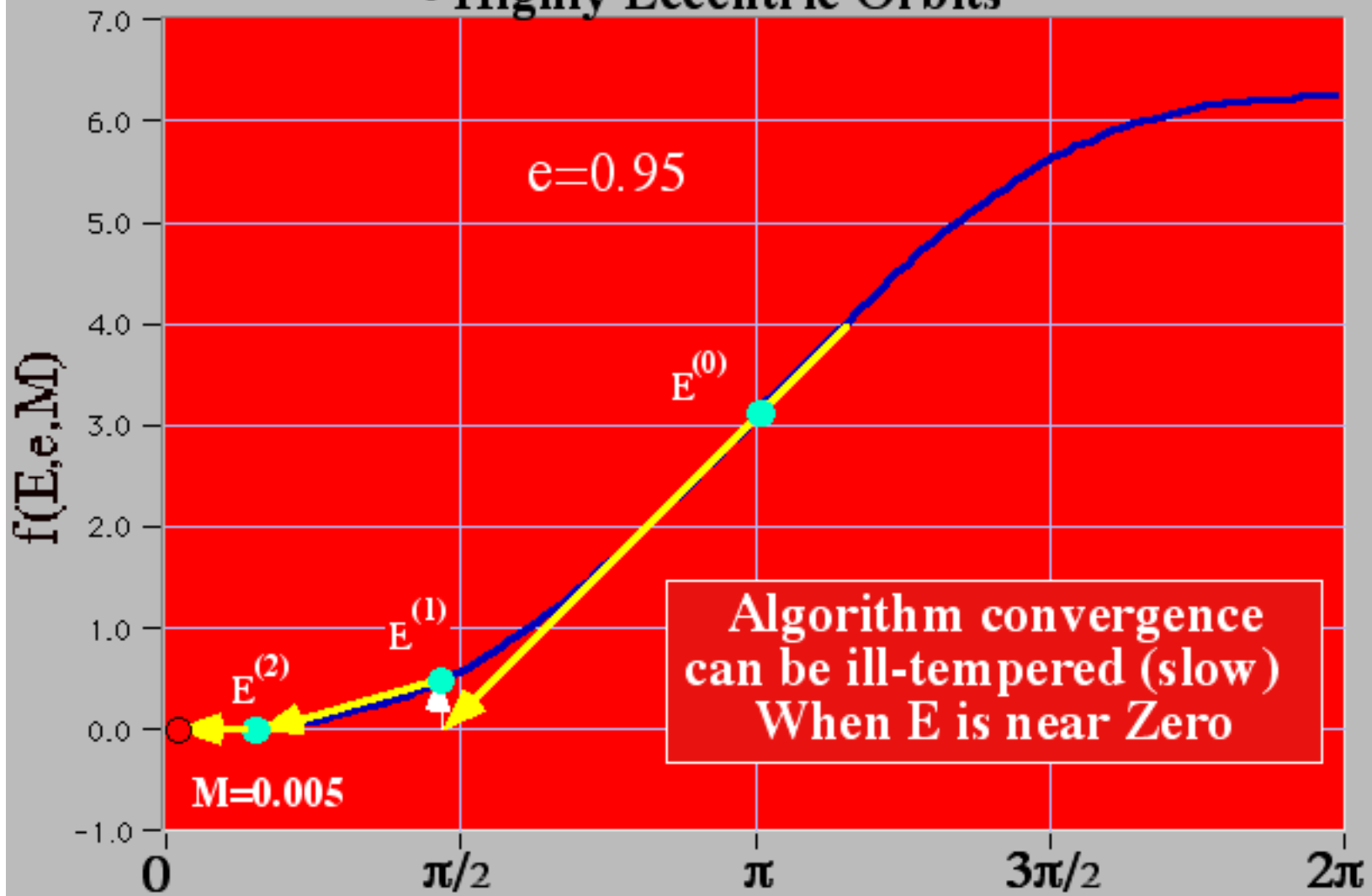
• Highly Eccentric Orbits



Eccentric Anomaly,  $E$  (radians)

# Starting Value (cont'd)

• Highly Eccentric Orbits



Eccentric Anomaly, E (radians)

## Starting Value (cont'd)

- Highly Eccentric Orbits

- Why is convergence "ill-tempered" (slow) near perigee?

$$E^{(j+1)} = E^{(j)} + \frac{M - [E^{(j)} - e \sin(E^{(j)})]}{1 - e \cos(E^{(j)})}$$

$$E^{(j+1)} = .11 + \frac{0.0051583 - [.11 - 0.95 \sin(.11)]}{1 - 0.95 \cos(.11)} =$$

$$0.11 + \frac{-0.0005523}{.05574} = .10009154$$

## Starting Value (cont'd)

- Highly Eccentric Orbits

Why is convergence "ill-tempered"  
(slow) near perigee?

- Normalized Change convergence criterion

$$\left| \frac{E^{(j+1)} - E^{(j)}}{E^{(j+1)} + E^{(j)}} \right| = 2 \left| \frac{.10009154 - 0.11}{.10009154 + 0.11} \right| = .0943$$

**only a 9.4% change**

... even though we are only 0.00009154 radians (.0052°) off  
in our estimate



## Starting Value (concluded)

- Since we are solving Kepler's Equation for each and every Time point in our propagation of the orbit ... convergence speed becomes critically important

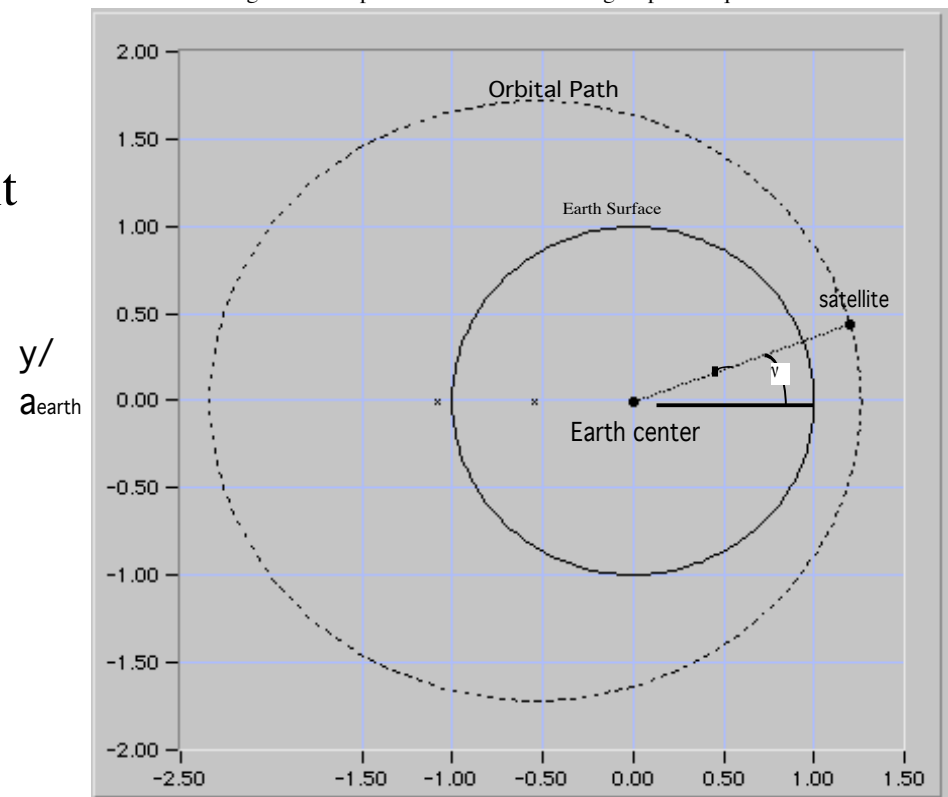
- Clearly, since convergence near the Perigee of an eccentric orbit can be a bit Of a problem ...

Its clear that we need .....

- a better way to start each iteration

- Next time  
... "Startup Algorithms"

Figure 5: Sample Orbit Calculation using Kepler's Equation



## OK ... so we need a better startup algorithm

- So where do we start?... with a really Simple ad-hoc solution .....
- When  $M \sim 0$ , we simply “kick it off zero” by adding or subtracting ...  $e$  (**Vallado Algorithm**)

$$M = E - e \sin(E)$$

$$\Downarrow$$

$$\text{Startup} \quad \left\{ \begin{array}{l} 0 \leq M \leq \pi: E_0 = M + e \\ \pi < M < 2\pi: E_0 = M - e \end{array} \right.$$

$$\text{Method 1} \quad \left\{ \begin{array}{l} 0 \leq M \leq \pi: E_0 = M + e \\ \pi < M < 2\pi: E_0 = M - e \end{array} \right.$$

# Derivation of Vallado Startup Algorithm\*

- Regroup Kepler's Equation and Expand in a Taylor's series

$$M = E - e \sin [E] \Rightarrow E(M) = M + e \sin [E]$$

$$E(M) = E(0) + M \left[ \frac{\partial [M + e \sin [E]]}{\partial M} \right]_0 + O(M^2)$$

- But

∇

$$\text{when } M = 0 \Rightarrow E(0) = e \sin [E(0)] \Rightarrow E(0) = 0$$

\* Proof is Courtesy of Brian M. Moore  
CPT, US Army, Naval Postgraduate School  
Space Systems Engineering, bmmoore@nps.navy.mil

# Derivation of Vallado Startup Algorithm\*(cont'd)

- and the Taylor's series reduces to

$$E(M) = M \left[ 1 + e \cos[E] \frac{\partial E}{\partial M} \right]_0 + O(M^2)$$

- But

$$\frac{\partial [M + e \sin [E]]}{\partial M} = \frac{\partial E}{\partial M} \Rightarrow \frac{\partial E}{\partial M} = 1 + e \cos[E] \frac{\partial E}{\partial M}$$

$$\frac{\partial E}{\partial M} = \frac{1}{1 - e \cos[E]} \Rightarrow \boxed{1 + e \cos[E] \frac{\partial E}{\partial M} = 1 + \frac{e \cos[E]}{[1 - e \cos[E]]}}$$

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# Derivation of Vallado Startup Algorithm\*(cont'd)

- Solving for the derivative

$$\frac{\partial E}{\partial M} = \frac{1}{1 - e \cos[E]} \Rightarrow \boxed{1 + e \cos[E] \frac{\partial E}{\partial M} = 1 + \frac{e \cos[E]}{[1 - e \cos[E]]}}$$

- and the Taylor's series further reduces to

$$E = M \left[ 1 + \frac{e \cos[E]}{[1 - e \cos[E]]} \right]_0 + O(M^2)$$

$$E = M \left[ 1 + \frac{e}{[1 - e]} \right] + O(M^2)$$

\* Proof is Courtesy of Brian M. Moore  
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# Derivation of Vallado Startup Algorithm\*(cont'd)

- Truncating the series after first order

$$\hat{E} \approx M + e \left[ \frac{M}{[1-e]} \right] \Rightarrow \left[ \begin{array}{l} 0 \leq M < \pi \Rightarrow \frac{M}{[1-e]} > 0 \\ -\pi \leq M < 0^{(-)} \Rightarrow \frac{M}{[1-e]} < 0 \end{array} \right]$$

- and now the “*Vallado approximation*”

$$\boxed{\left| \frac{M}{[1-e]} \right| \approx 1} \Rightarrow \left[ \begin{array}{l} 0 \leq M < \pi \Rightarrow \hat{E} \approx M + e \\ -\pi \leq M < 0^{(-)} \Rightarrow \hat{E} \approx M - e \end{array} \right]$$

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# Derivation of Vallado Startup Algorithm\*(cont'd)

- “*Vallado approximation*”

$$\left| \frac{M}{[1-e]} \right| \approx 1$$

- Startup is most accurate for  $M \sim 1-e \rightarrow$  for highly eccentric Orbits ... this value for M also happens to be near Perigee
- For low eccentricity orbits ... we've already seen that It really doesn't matter where you start
- But how about intermediate eccentricity orbits, I.e.  $e=0.5$   
 $M \rightarrow 1-e = 0.5$  radians  $\sim 28.65^\circ$  ... ισ τηισ γοοδ ενουγη?

\* Proof is Courtesy of Brian M. Moore  
CPT, US Army, Naval Postgraduate School  
Space Systems Engineering, bmmoore@nps.navy.mil

# Linear Interpolation as a Startup Algorithm

$$M_{t-0} = \left\{ E_t - e \sin(E_t) \right\}$$

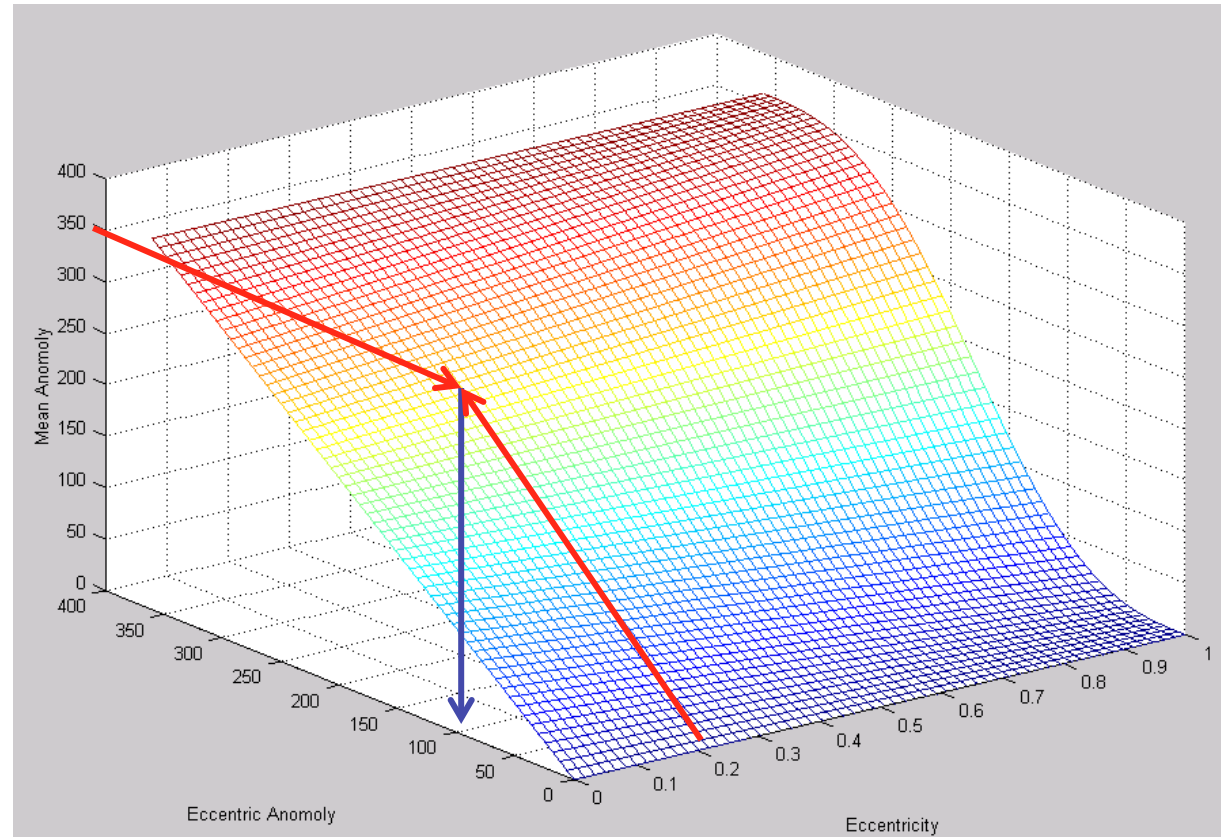
- Exploit the fact that Kepler's equation is *explicit* in the “forward direction”

\* Proof is Courtesy of Brian M. Moore  
CPT, US Army, Naval Postgraduate School  
Space Systems Engineering, bmmoore@nps.navy.mil



# Linear Interpolation as a Startup Algorithm (cont'd)

- Generate 3-D Space of  $\{M, e, E\}$  that Satisfy Kepler's Equation
- Use 2-D Interpolation Of  $\{M, e, E\}$  To Generate Starting Guess



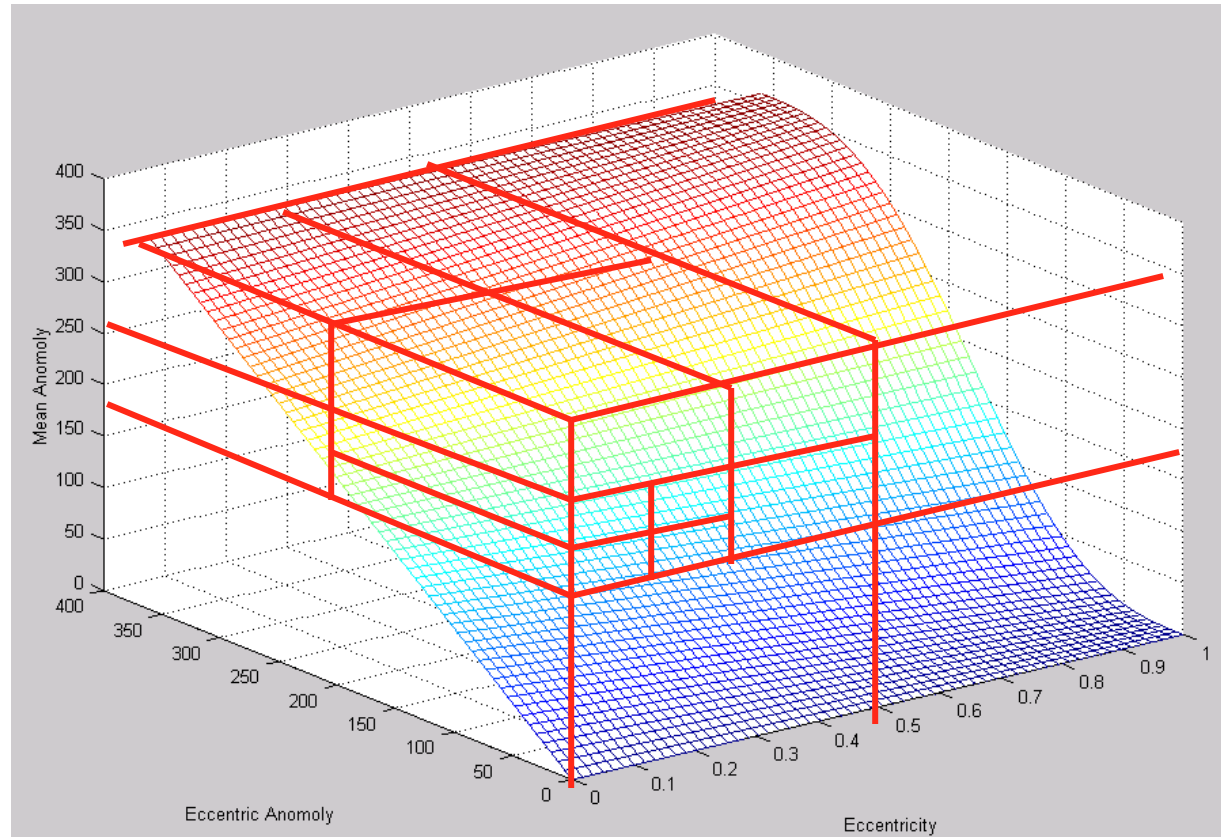
\* Proof is Courtesy of Brian M. Moore  
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# Linear Interpolation as a Startup Algorithm (cont'd)

- Using a “quaternary” Search pattern Interpolation is fast

- But stored Data base is Rather large

- Mesh Mesh Density fixed By maximum Eccentricity of Orbits to be analyzed



\* Proof is Courtesy of Brian M. Moore  
CPT, US Army, Naval Postgraduate School  
Space Systems Engineering, [bmmoore@nps.navy.mil](mailto:bmmoore@nps.navy.mil)

## OK ... can we develop another startup algorithm?

- So where do we start?... with the *Fourier series solution of Kepler's Equation* ....\*

\*Dörrie, H. "The Kepler Equation." 100 Great Problems of Elementary Mathematics: Their History and Solutions. New York: Dover, pp. 330-334, 1965.

## Fourier Series Solution

$$M = E - e \sin(E)$$



$$E = M + \sum_{n=1}^{\infty} \left[ \frac{2}{n} J_n(n e) \sin(n M) \right]$$

$$J_n(n e) = \sum_{j=0}^{\infty} \frac{(-1)^j}{j! (n+j)!} \left( \frac{n e}{2} \right)^{n+2j} \Rightarrow \left\{ \begin{array}{l} \text{Bessel Function} \\ \text{of the First Kind} \end{array} \right.$$

- Very inefficient way to solve Kepler's equation

## Fourier Series Solution (cont'd)

- Buuuutt ... if we expand out the terms in the series

$$E \approx M + \left[ 2 J_1(e) \sin(M) \right] + \left[ J_2(2e) \sin(2 M) \right] + \left[ \frac{2}{3} J_3(3e) \sin(3 M) \right] + \dots$$

$$J_1(e) \approx \frac{e}{2} - \frac{e^3}{16} + \frac{e^5}{384} + \dots$$

$$J_2(e) \approx \frac{e^2}{2} - \frac{e^4}{6} + \frac{e^6}{48} + \dots$$

$$J_3(e) \approx \frac{9 e^3}{16} - \frac{81 e^5}{256} + \frac{729 e^7}{10240} + \dots$$

## Fourier Series Solution (cont'd)

- Substituting in for the  $J_i$ 's and collecting terms:

$$E \approx M + \left( e - \frac{e^3}{8} + \frac{e^5}{192} \right) \sin ( M ) +$$
$$\left( \frac{e^2}{2} - \frac{e^4}{6} + \dots \right) \sin ( 2 M ) +$$
$$\left( \frac{3 e^3}{8} - \frac{27e^5}{128} + \dots \right) \sin ( 3 M ) + \dots$$

## Near the Troublesome point at perigee

- at perigee ...  $\sin(\mu M) \sim \mu M, \{ \mu = 0, 1, 2, \dots \}$

$$E_0 \approx M + \left( e - \frac{e^3}{8} + \dots \right) M +$$

$$\left( \frac{e^2}{2} - \frac{e^4}{6} + \dots \right) 2 M +$$

$$\left( \frac{3 e^3}{8} - \frac{27 e^5}{128} + \dots \right) 3 M + \dots \approx$$

$$M [1 + e + e^2 + e^3 + \dots]$$

# Fourier Series Method Near Perigee

Startup  
Method 2

$$E_0 \approx M [1 + e + e^2 + e^3]$$

- Higher Order Method ... but still simple to implement



## Taylor Series Method

- Kepler's Equation

$$E - e \sin(E) - M = 0$$

- Expand "Sine" term in Taylor's series

$$\sin(E) = E - \frac{E^3}{3!} + \frac{E^5}{5!} \dots$$


- Truncate : **At** '5th" Order term

Fourth-order accurate

## Taylor Series Method (cont'd)

- Substitute Truncated Taylor's Series into Kepler's Equation

$$\hat{E}^{(0)} - e \left[ \hat{E}^{(0)} - \frac{(\hat{E}^{(0)})^3}{3!} \right] - M = 0$$

- Solve for  $E^3$  term ... to give startup algorithm

$$(\hat{E}^0)^3 + \frac{6(1-e)}{e} \hat{E}^0 - \frac{6M}{e} = 0$$

## Taylor Series Method (cont'd)

$$(\hat{E}^0)^3 + \frac{6(1-e)}{e} \hat{E}^0 - \frac{6M}{e} = 0$$

- "Resolvent Cubic" Equation has Closed-form Solution  
... two complex roots (complex root 1)

$$\frac{3 * 2^{1/3} (+.j \sqrt{3}) (-1 + e)}{(162 e^2 M + \sqrt{-23328 (-1 + e)^3 e^3 + 26244 e^4 M^2})^{1/3}} - \frac{(1 + j \sqrt{3}) (162 e^2 M + \sqrt{-23328 (-1 + e)^3 e^3 + 26244 e^4 M^2})^{1/3}}{6 * 2^{1/3} e}$$

## Taylor Series Method (cont'd)

$$(\hat{E}^0)^3 + \frac{6(1-e)}{e} \hat{E}^0 - \frac{6M}{e} = 0$$

- "Resolvent Cubic" Equation has Closed-form Solution  
.... two complex roots (complex root 2)

$$\frac{3 * 2^{1/3} (1 - j \sqrt{3}) (-1 + e)}{(162 e^2 M + \sqrt{-23328 (-1 + e)^3 e^3 + 26244 e^4 M^2})^{1/3}} - \frac{(1 + j \sqrt{3}) (162 e^2 M + \sqrt{-23328 (-1 + e)^3 e^3 + 26244 e^4 M^2})^{1/3}}{6 * 2^{1/3} e}$$

## Taylor Series Method (cont'd)

$$(\hat{E}^0)^3 + \frac{6(1-e)}{e} \hat{E}^0 - \frac{6M}{e} = 0$$

- "Resolvent Cubic" Equation has Closed-form Solution

....and one REAL root ... which is the one we want ...

$$\hat{E}^{(0)} =$$

Startup Method 3:

$$\frac{-2e + 2e^2 + \left( 3e^2 M + \sqrt{e^3 (-8(-1+e)^3 + 9eM^2)} \right)^{2/3}}{e \left( 3e^2 M + \sqrt{e^3 (-8(-1+e)^3 + 9eM^2)} \right)^{1/3}}$$

## OK ... How Good are these Startup Values

- From Vallado (Ad-Hoc Solution)  
... the Traditional Startup Assumption is

$$\hat{E}^{(0)} = \begin{array}{l} M + e \quad (0 \leq M \leq \pi) \\ M - e \quad (\pi \leq M \leq 2\pi) \end{array}$$

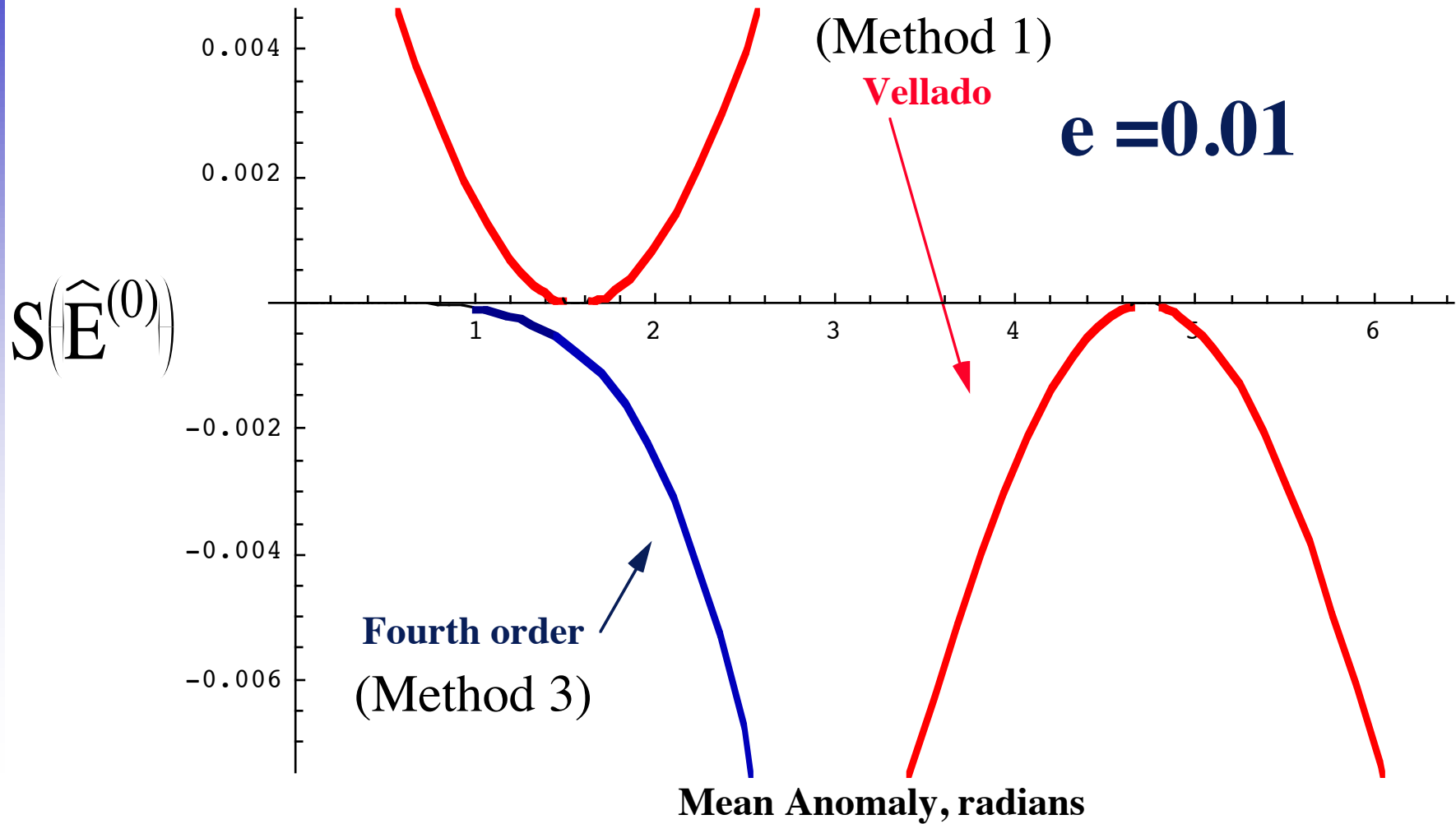
## Startup Accuracy Metric

- The Closer  $S(\hat{E}^{(0)})$  is to "zero", then the more accurate the startup value

$$S(\hat{E}^{(0)}) \equiv \hat{E}^{(0)} - e \sin(\hat{E}^{(0)}) - M$$

$$\text{when } \hat{E}^{(0)} = E \Rightarrow S(\hat{E}^{(0)}) = 0$$

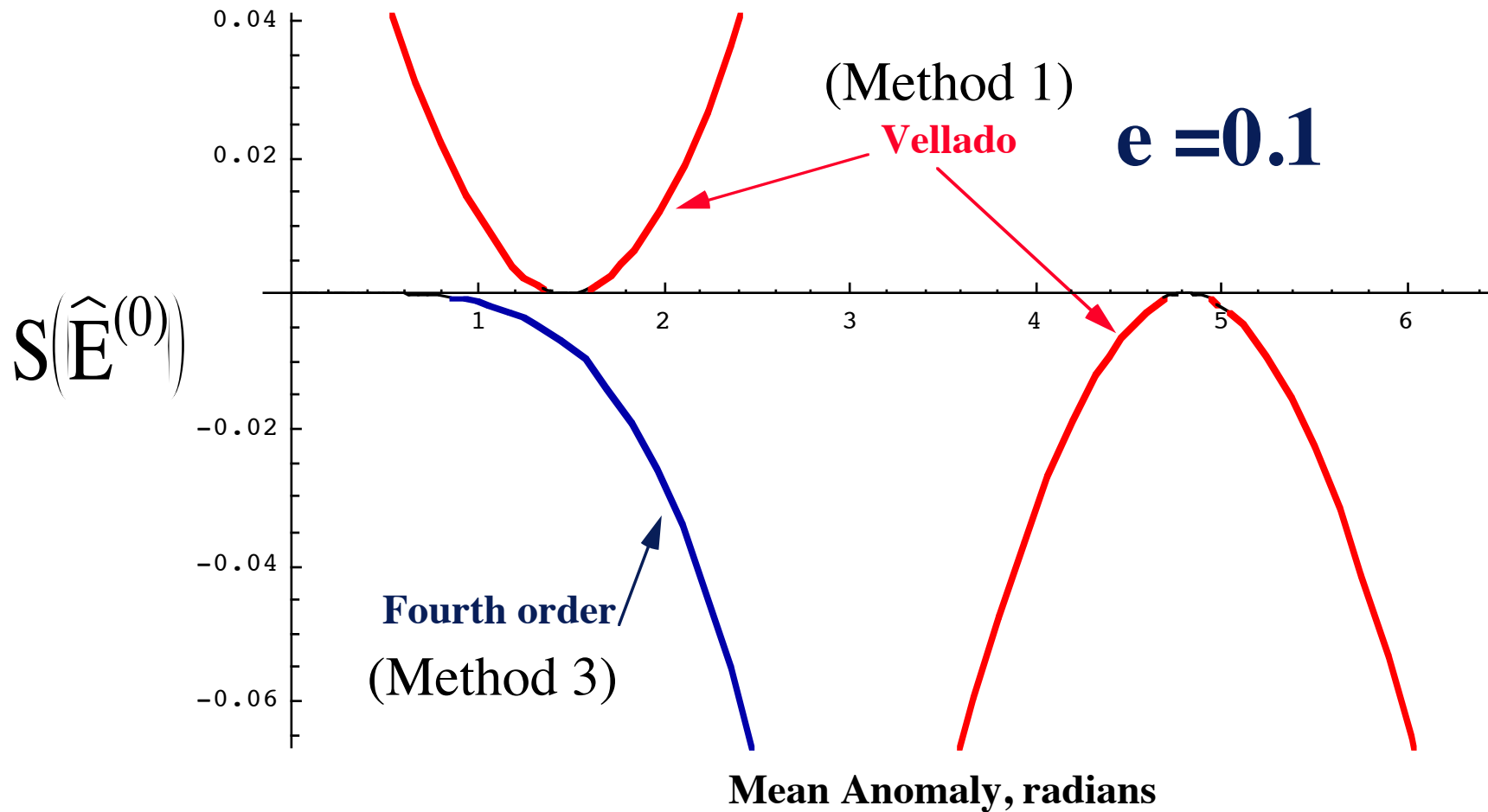
# Comparison of Start-up Values





# Comparison of Start-up Values

(cont'd)



# Comparison of Start-up Values

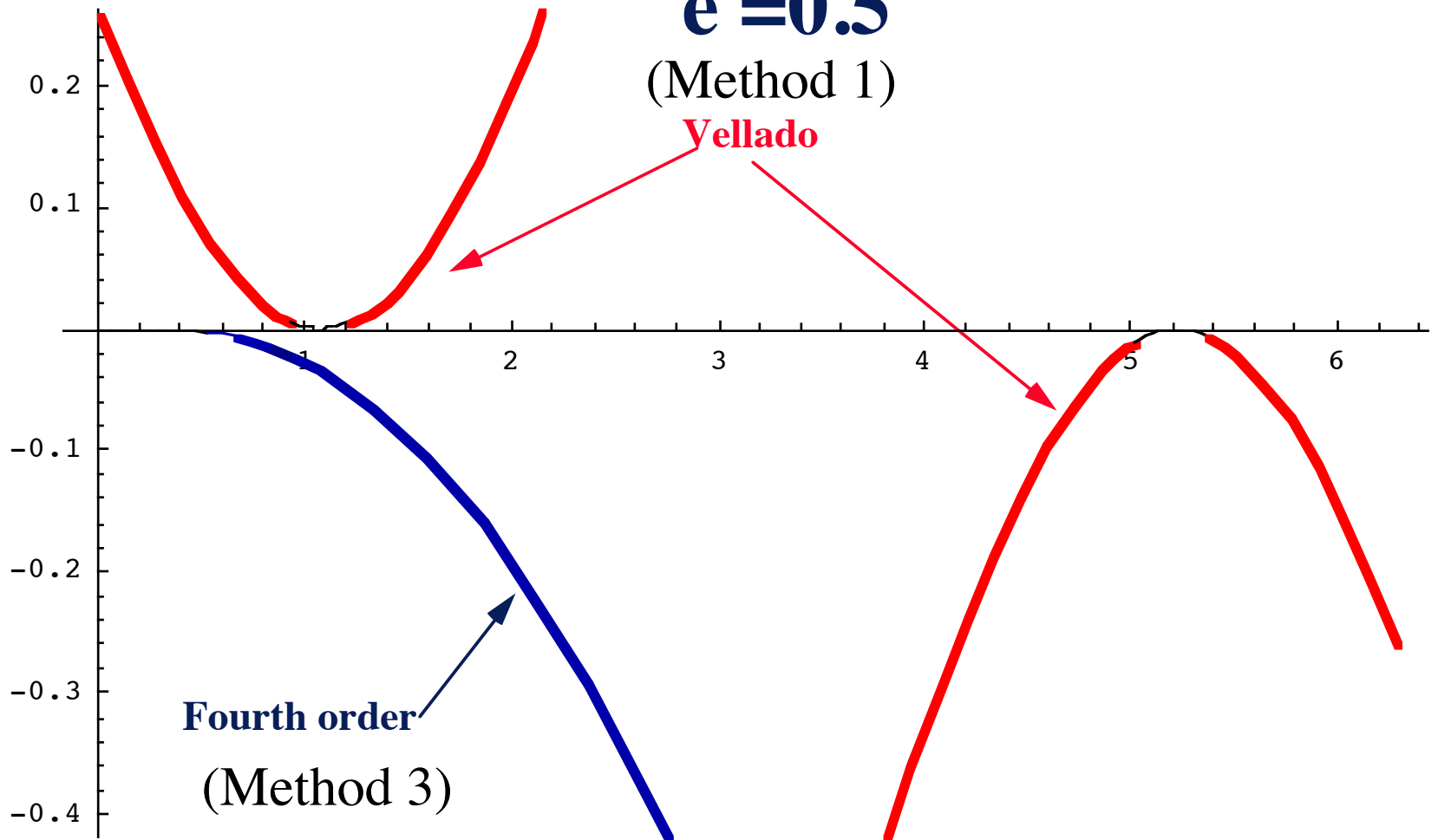
(cont'd)

$e = 0.5$

(Method 1)

Yellado

$S(\hat{E}^{(0)})$



Fourth order  
(Method 3)

Mean Anomaly, radians

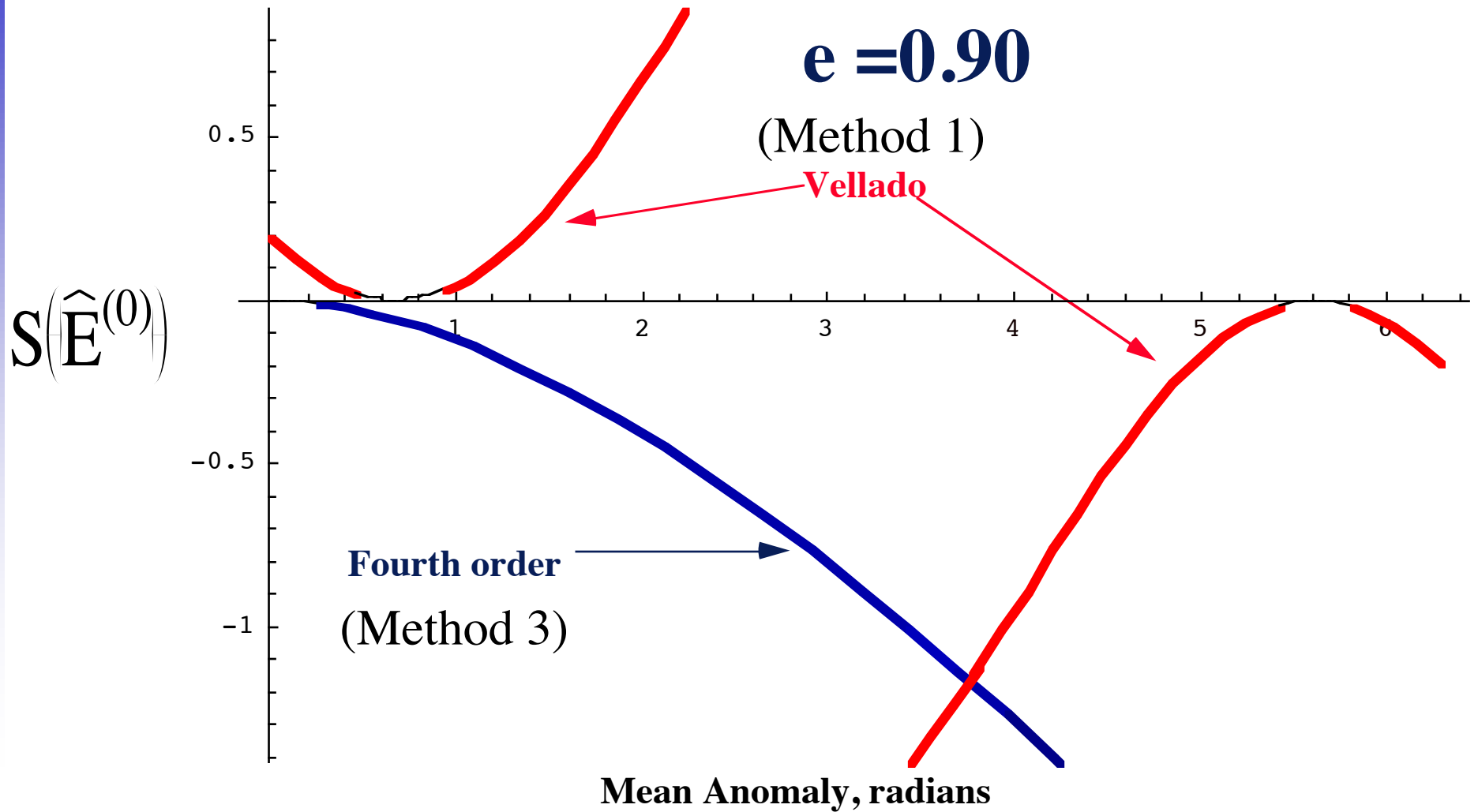
# Comparison of Start-up Values

(cont'd)

$e = 0.90$

(Method 1)

Vellado



# Iteration Plots (e=0.01)

#of points

100

eccentricity

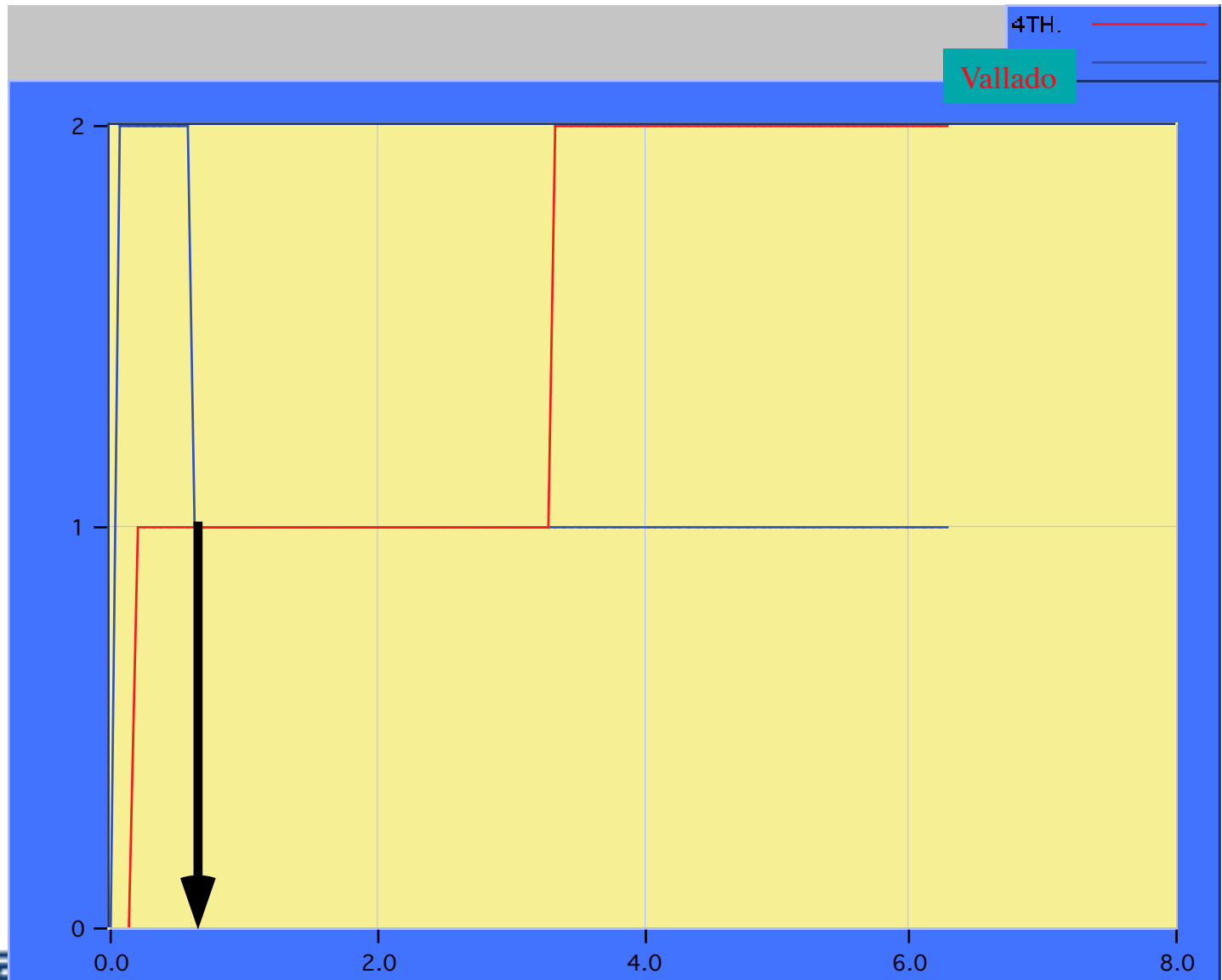
0.01

Max # of Iterations

25

% error required

0.0000100



# Utah State UNIVERSITY Iteration Plots (e=0.1)

#of points

↔ 100

eccentricity

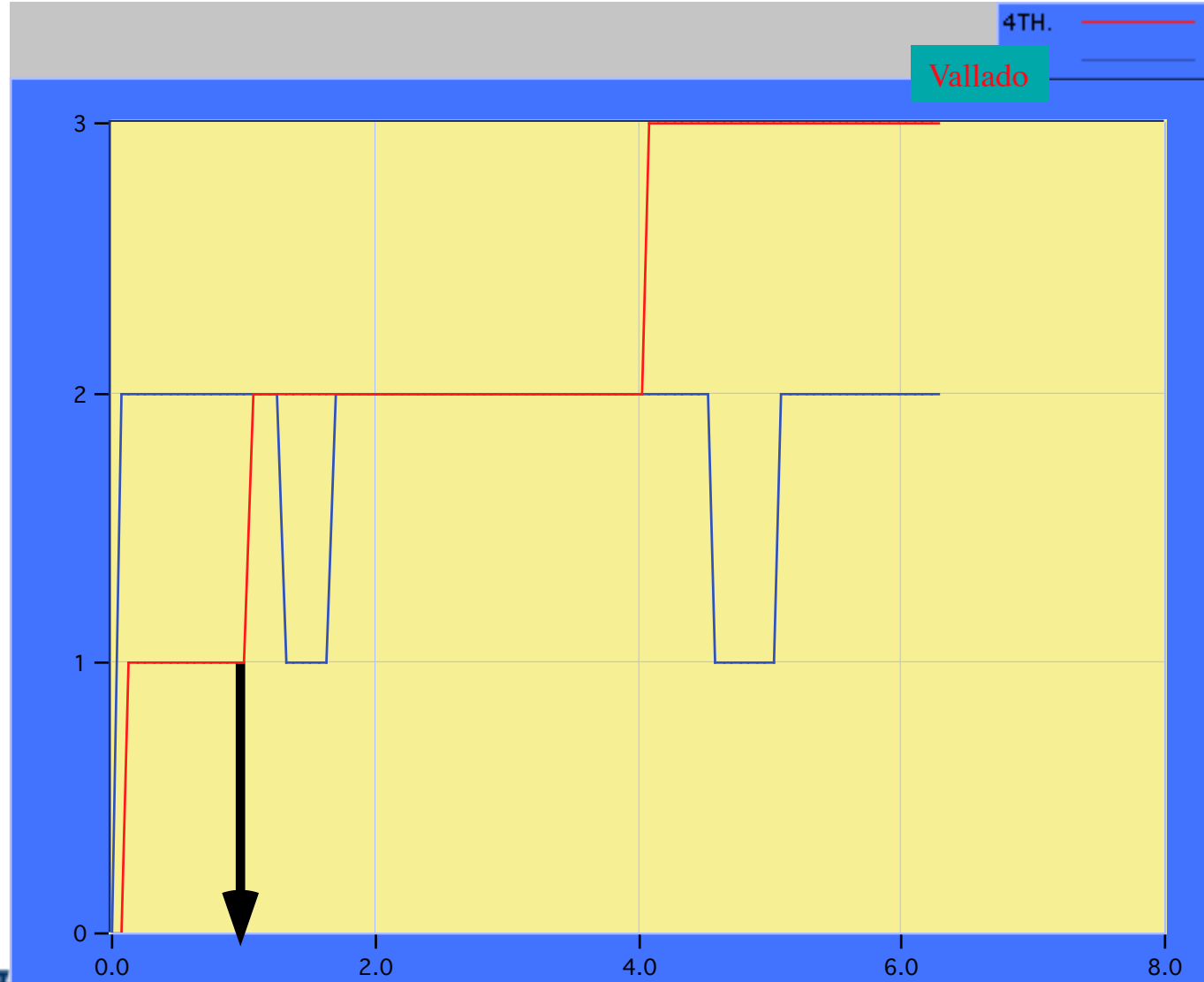
↔ 0.10

Max # of Iterations

↔ 25

% error required

↔ 0.0000100



# Iteration Plots (e=0.5)

#of points

100

eccentricity

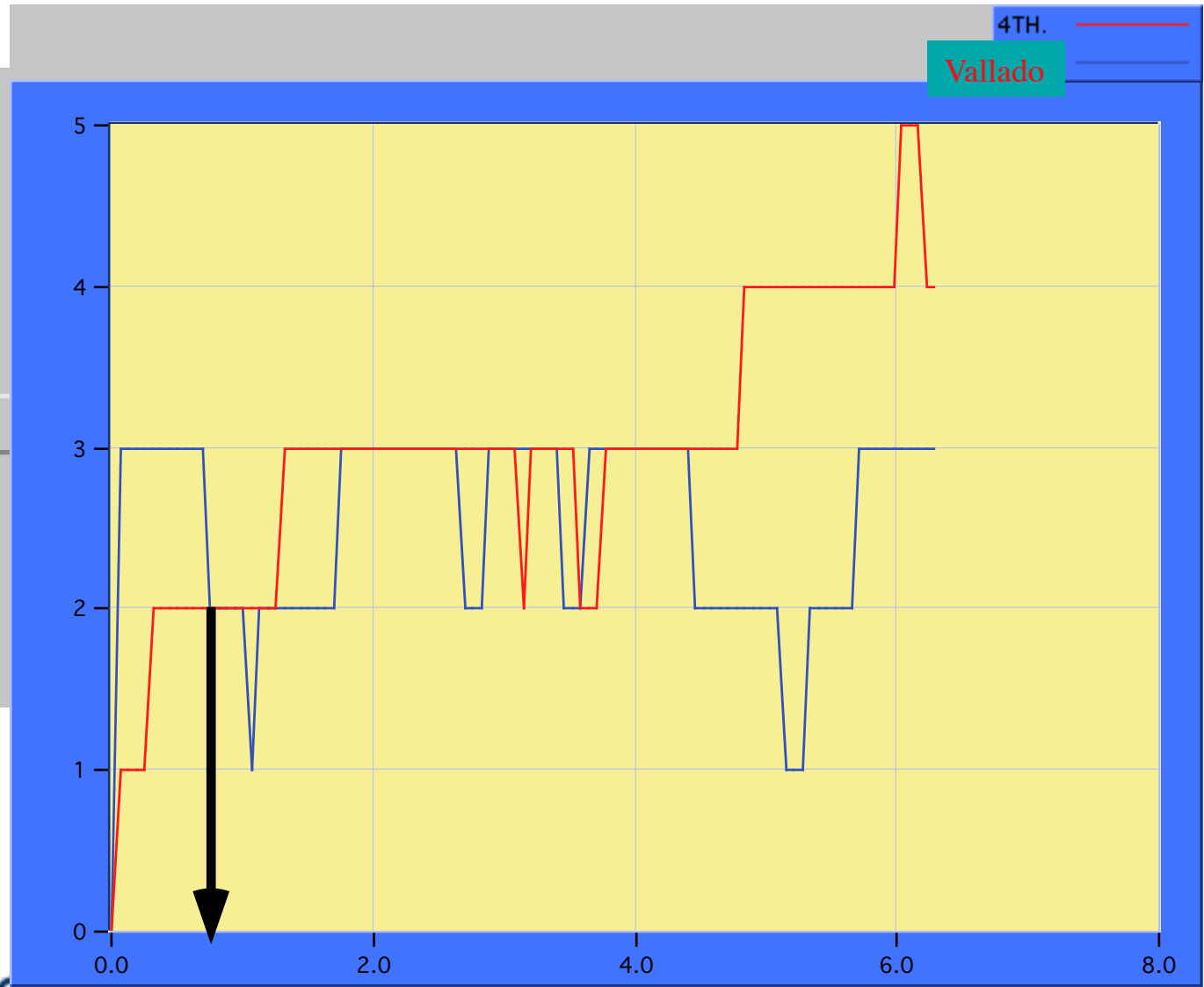
0.50

Max # of Iterations

25

% error required

0.0000100





# Iteration Plots (e=0.9)

#of points

↔ 100

eccentricity

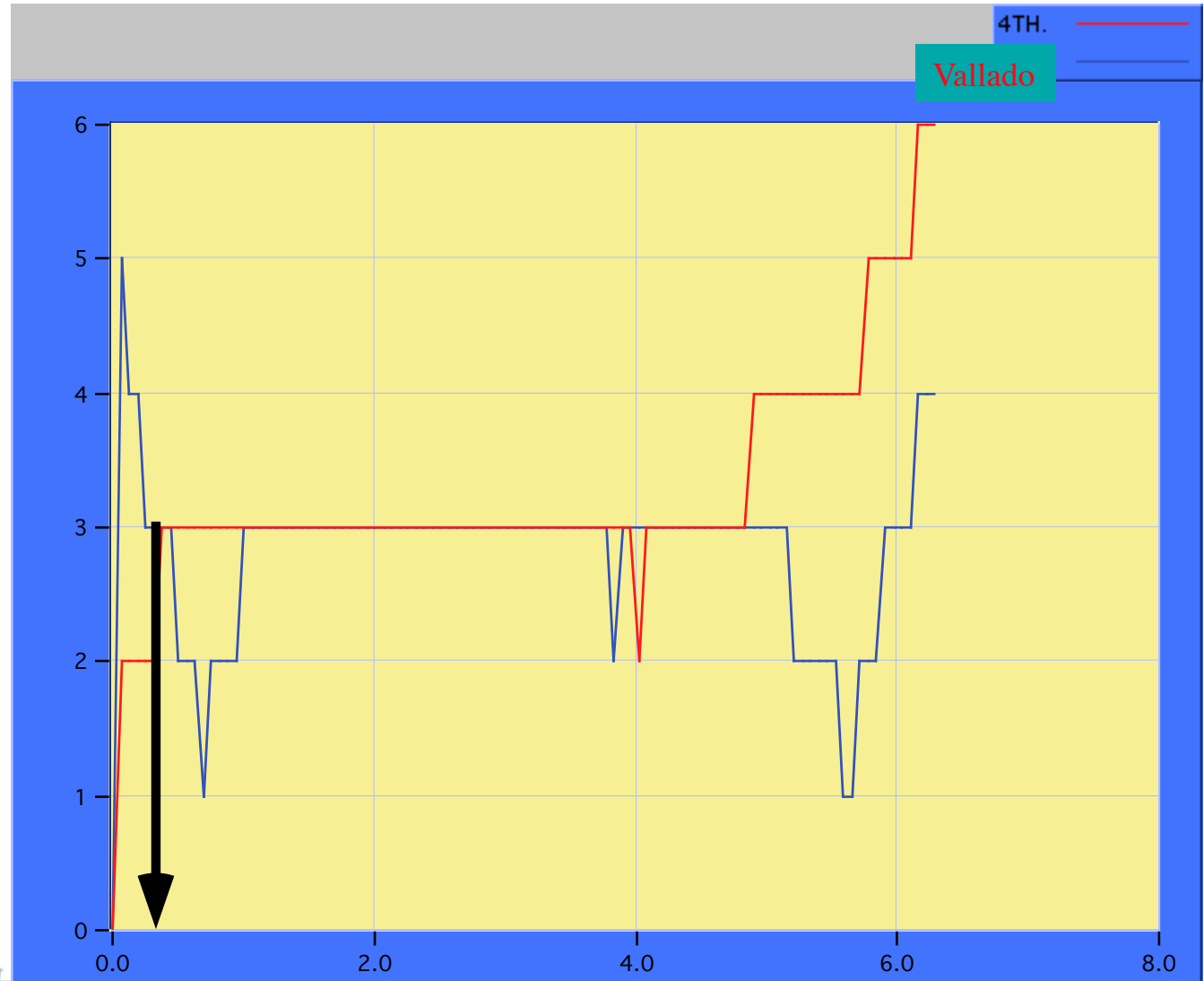
↔ 0.90

Max # of Iterations

↔ 25

% error required

↔ 0.0000100



## Conclusions?

- **For Most Conditions the Simple "Vallado" Startup Method gives Superior convergence**  
... simplicity of the algorithm clearly justified its use
- **Near the perigee, the 4th order (Taylor's series) startup gives better convergence**
- **Where's the "push" point ... it appears that for all eccentricities the Taylor's serie startup method offers convergence advantages for**

$$v < 0.25 \text{radians } (\sim 15^\circ)$$

- *Gives a convergence aid for solutions near perigee*



**Soooo... its looks like**

**M > 0.25 radians ... use**

$$\hat{E}^{(0)} = \begin{cases} M + e & (0 \leq M \leq \pi) \\ M - e & (\pi \leq M \leq 2\pi) \end{cases}$$

**Otherwise .... use**

$$\hat{E}^{(0)} = \frac{-2e + 2e^2 + \left( 3e^2 M + \sqrt{e^3 (-8(-1+e)^3 + 9eM^2)} \right)^{2/3}}{e \left( 3e^2 M + \sqrt{e^3 (-8(-1+e)^3 + 9eM^2)} \right)^{1/3}}$$